



Conceptual design of the retroreflector, photodetector, and optical beacon payloads for the Photon target satellite

R. Glenn Sellar, Ronald L. Phillips, Larry. C. Andrews, Cynthia Y. Hopen,
Derek M. Shannon, Jennifer J. Huddle, Annabel Marcos, Jason Bachelor and Kiet Van
Florida Space Institute
12424 Research Parkway, Suite 400, Orlando, FL 32826
email: gsellar@mail.ucf.edu



* Funded by the Ballistic Missile Defense Organization, through the Space and Naval Warfare Systems Center San Diego, under contract # N66001-97-C-8644.





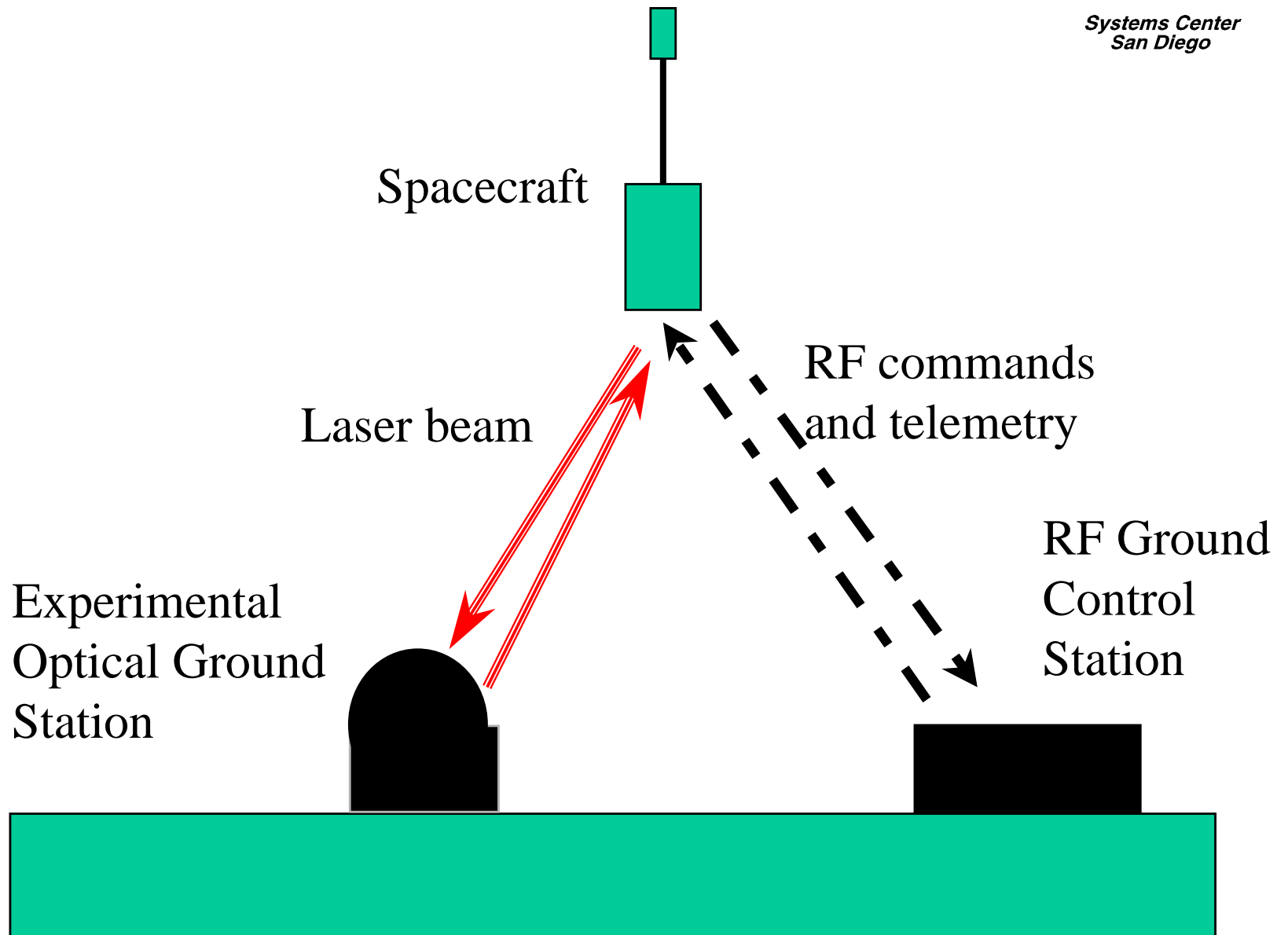
Retroreflectors in space



- Used for experiments in:
 - differential absorption lidar (DIAL)
 - atmospheric propagation
 - orbital tracking
 - laser radar
 - optical communications
 - etceteras ...
- Usually carried as secondary payloads
 - exceptions include LAGEOS, RIS



Mission Concept

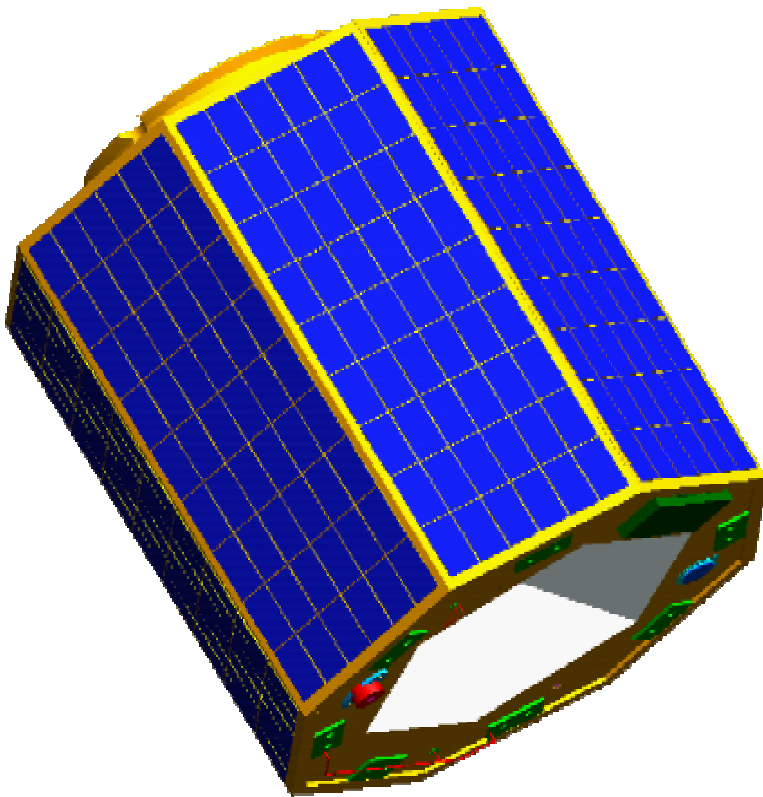




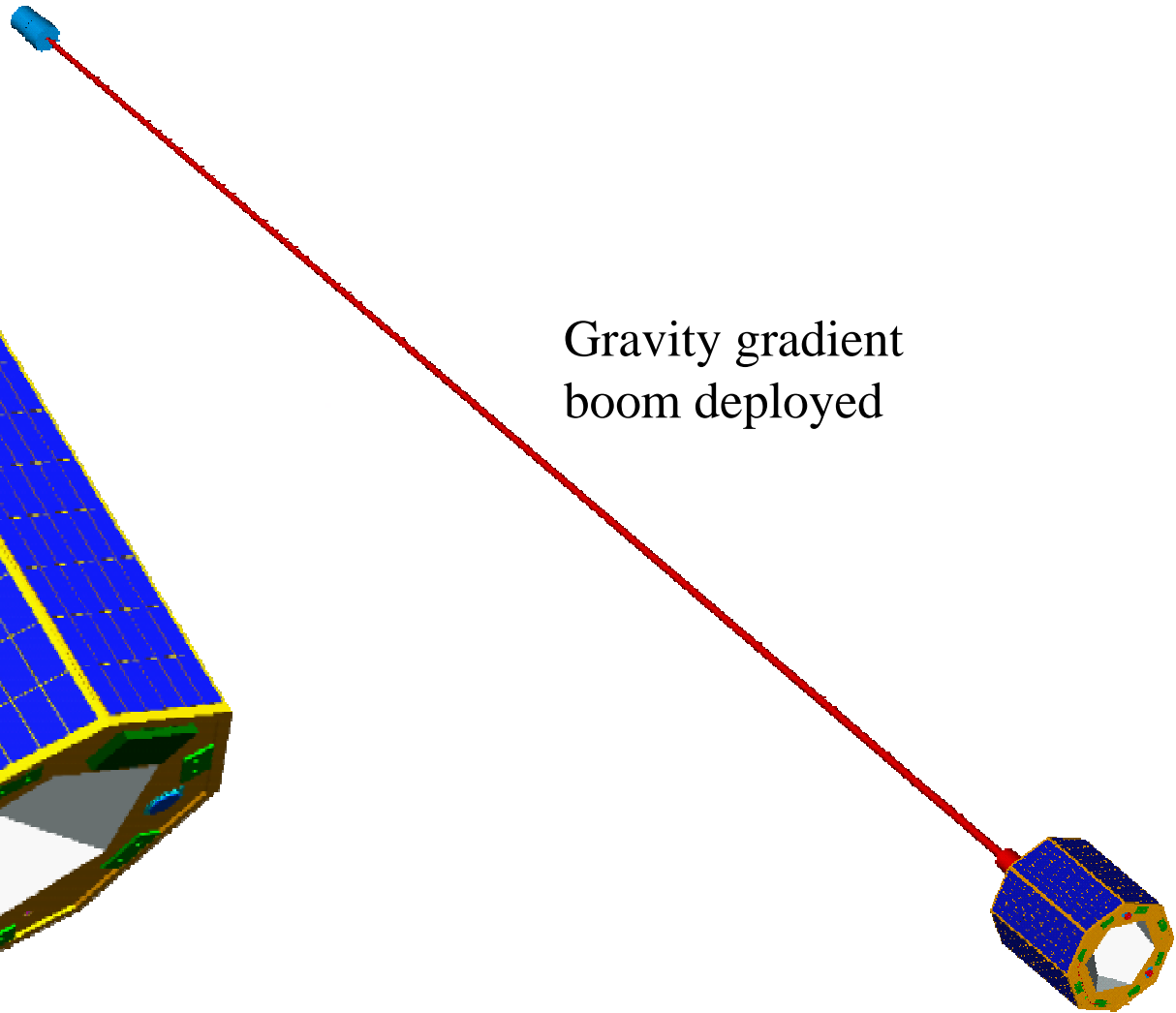
Photon Spacecraft



Gravity gradient
boom stowed



Gravity gradient
boom deployed





Primary Payloads



- Photon is a dedicated optical target spacecraft, carrying three payloads:
 - large (27 cm diameter) single retroreflector
 - optical beacon
 - photodetector



Photon Advantages



- Retroreflector:
 - complex/expensive equipment is located on the ground
 - accommodates a wide range of possible experiments
 - *single* retroreflector avoids interference effects exhibited by arrays
 - effective cross-section will exceed that of any others currently in orbit (or known to be in development)
- Photodetector:
 - provides independent data for the uplink path
- Optical beacon:
 - facilitates acquisition
 - provides independent information on the downlink path

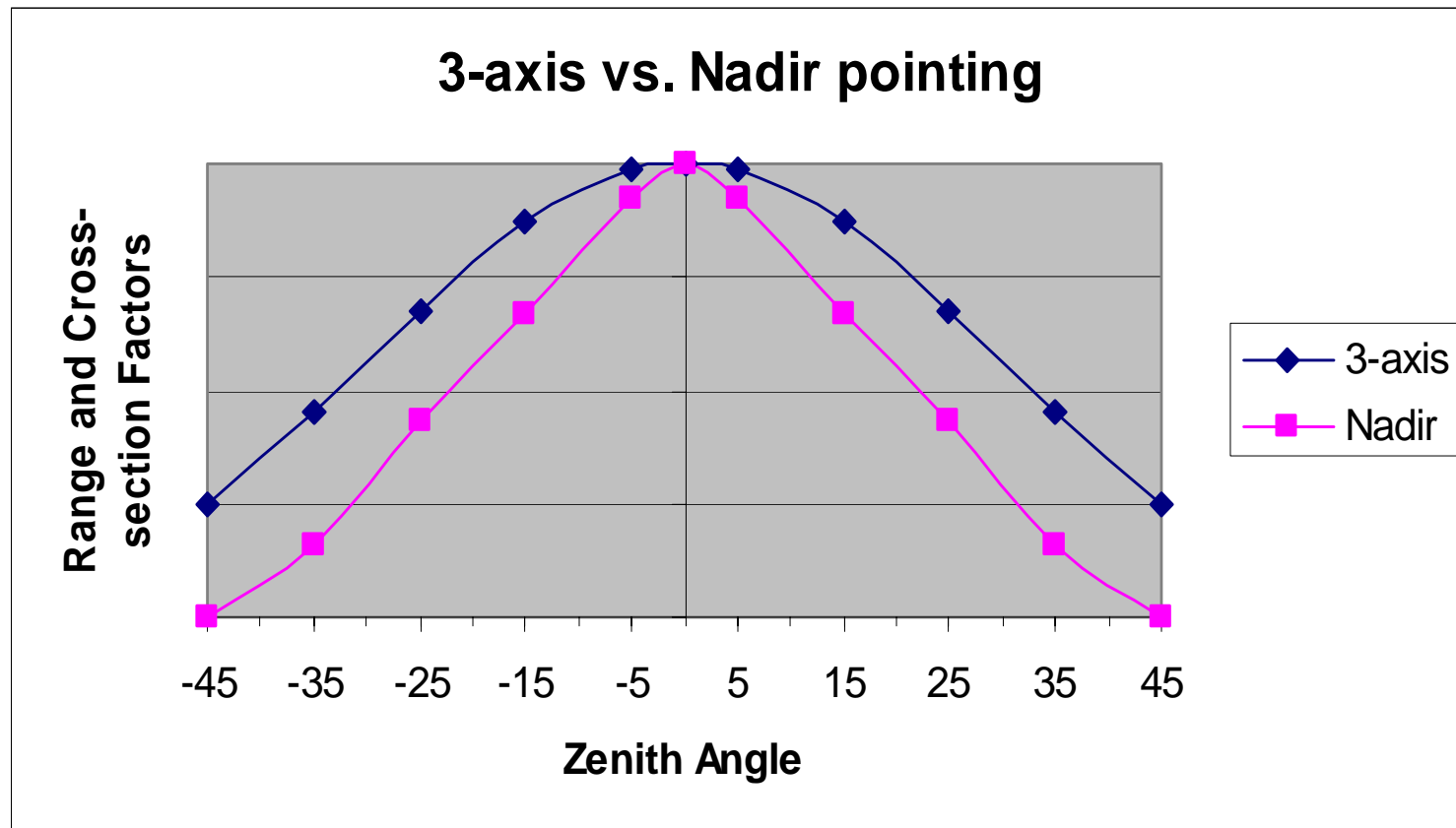


Comparison of spaceborne retroreflectors



Satellite	Minimum Range [km]	Effective Diameter [cm]	Figure of Merit D^2/R^4 [1/m ²]	Number of Retro's	Beacon	Detector	Notes
LAGEOS-1,-2	5860	24	2 E-28	426	NO	NO	
ADEOS	797	50	1 E-24	1	NO	NO	(failed)
AJSAT	1490	215	4 E-25	1436	NO	NO	
STARLETTE	812	24	6 E-25	60	NO	NO	
STELLA	800	24	6 E-25	60	NO	NO	
ERS-1,-2	780	18	5 E-25	9	NO	NO	
TOPEX/POSEIDON	1340	150	5 E-25	192	NO	NO	
METEOR-3	1180	28	1 E-25	24	NO	NO	
GFZ-1	396	20	8 E-24	60	NO	NO	
TIPS	1022	?	?	18	NO	NO	
GPS-35,-36	20200	24	1 E-30	?	NO	NO	
ETALON-1	19120	129	1 E-29	2134	NO	NO	
GLONASS	19140	120	9 E-30	396	NO	NO	
FIZEAU	931	15	2 E-25	3	NO	NO	
RESURS-01	678	15	7 E-25	1	NO	NO	
MSTI-II	431	18	5 E-24	9	NO	NO	
BE-C	940	?	?	? (multi)	NO	NO	
PHOTON	350	27	1.8 E-23	1	YES	YES	(planned)

Attitude control trade study





Predicted number of passes with elev.>60°

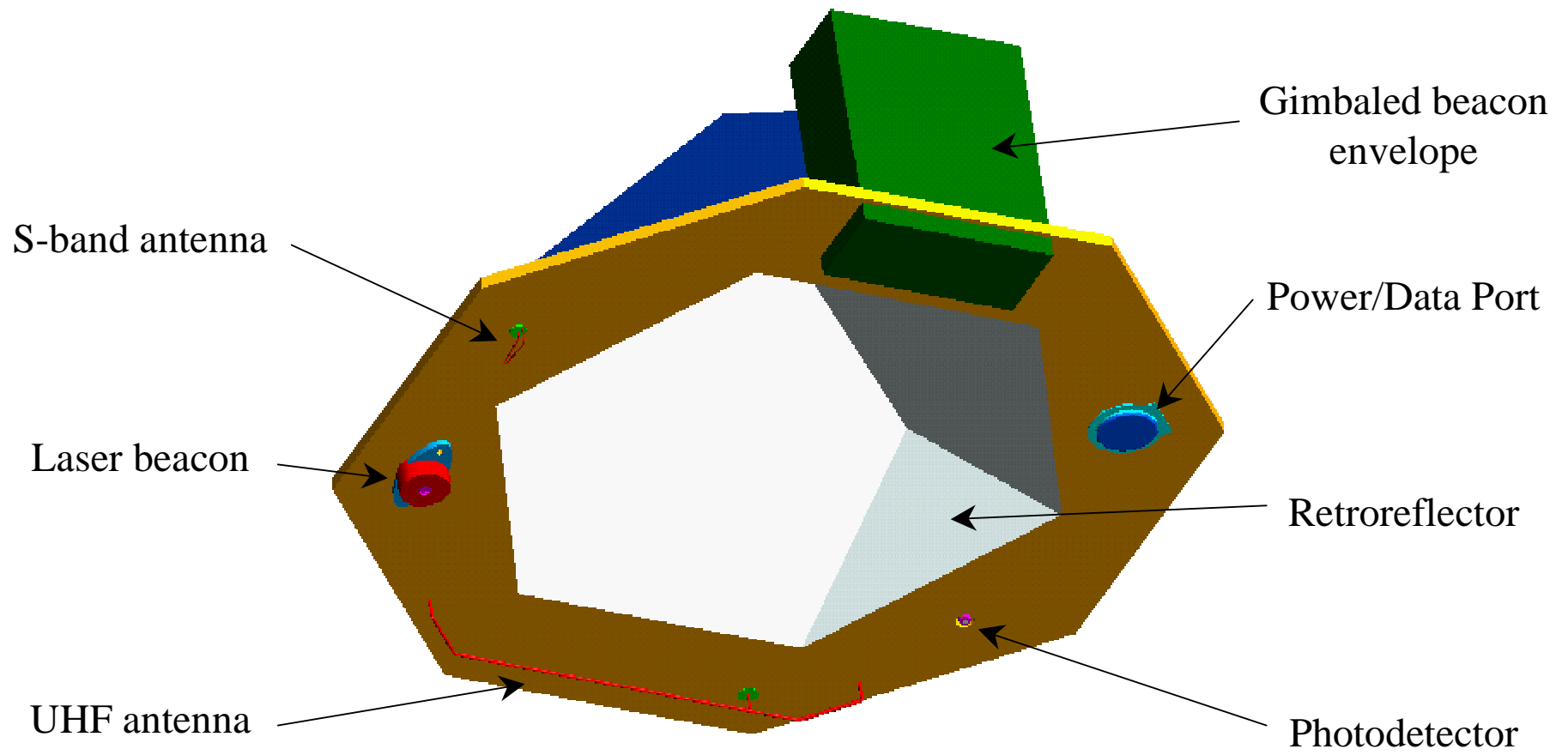


Orbit	Columbia	Space Station
Nominal Lifetime [days]	162	474
95% Prob. Lifetime [days]	85	241
Passes/year	706	177
Nominal Passes	313	230
95% Prob. Passes	164	117

- lifetimes based on 1/1/2001 launch date
- for ground station at 28.5° latitude

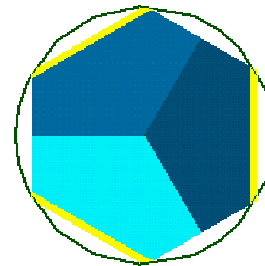
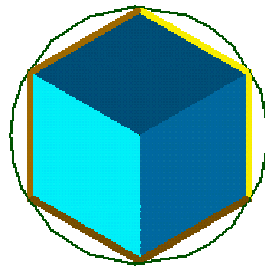
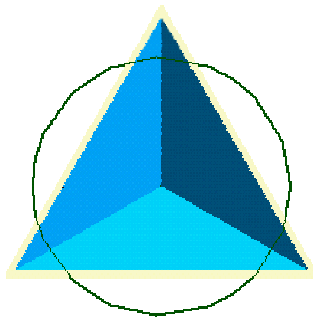


Payload Module



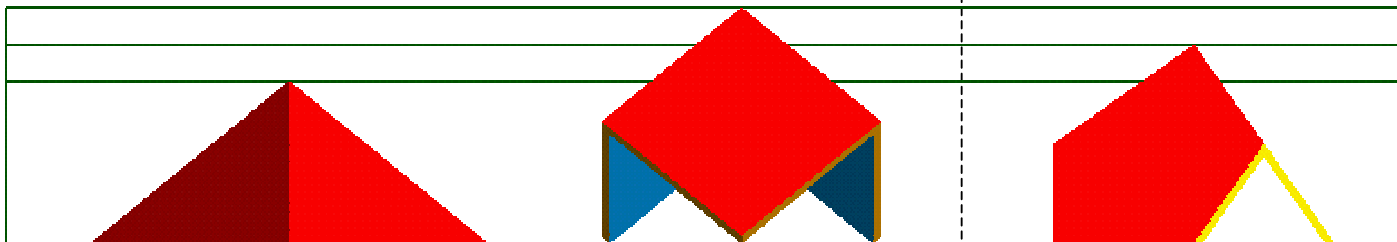
Retroreflector Design

Bottom View



Front View

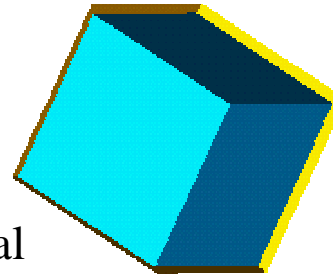
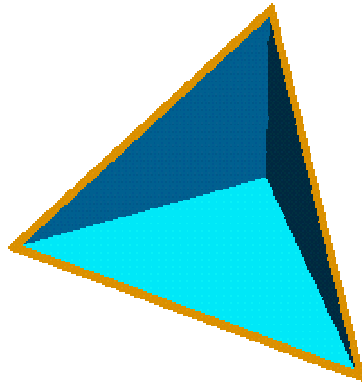
Maximum allowable radius



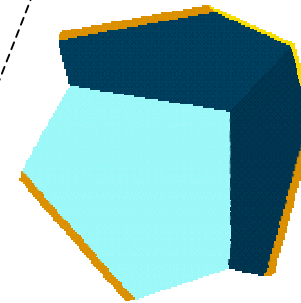
- Photon Retroreflector
 - Maximized cross sectional area
 - Minimized height

Retroreflector Design

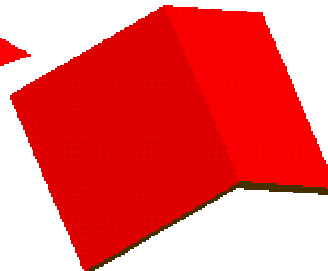
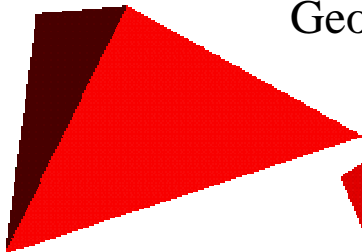
Isometric
Bottom View



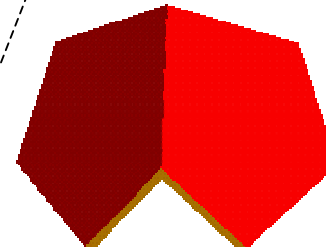
Traditional
Retroreflector
Geometry



Isometric
Top View

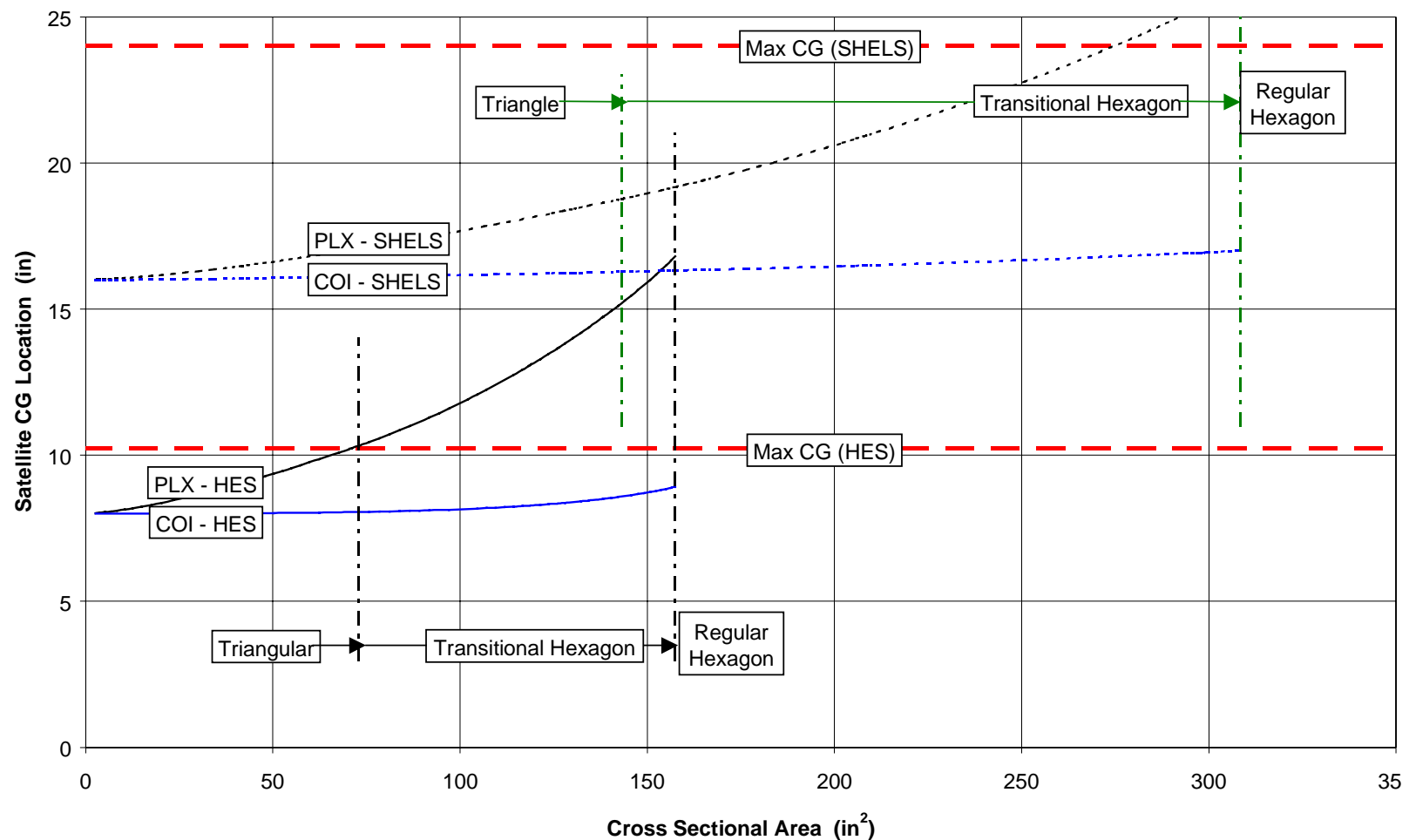


Photon
Retroreflector
Geometry

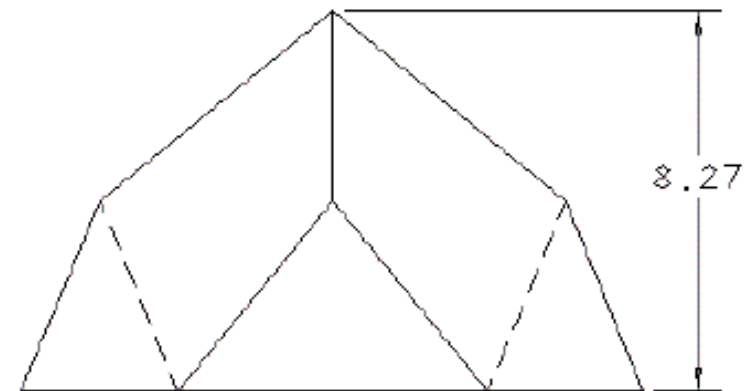
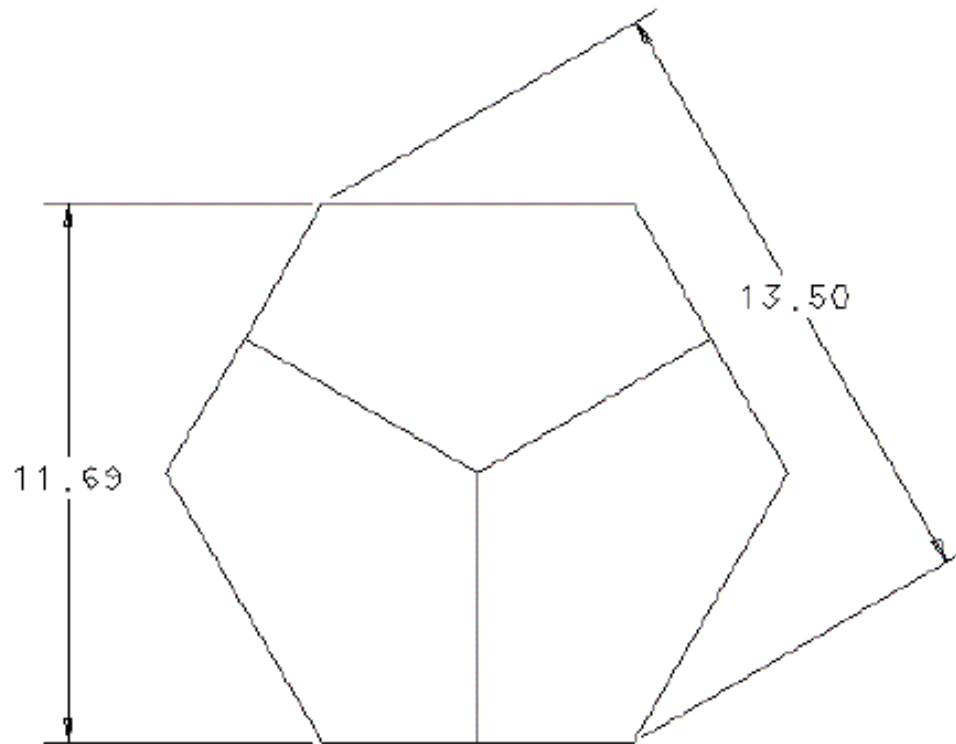


Retroreflector size constraints

Retroreflector Size vs. CG Location
CG of Satellite w/o Retro = 8.00 in (HES), 16.00 in (SHELS)

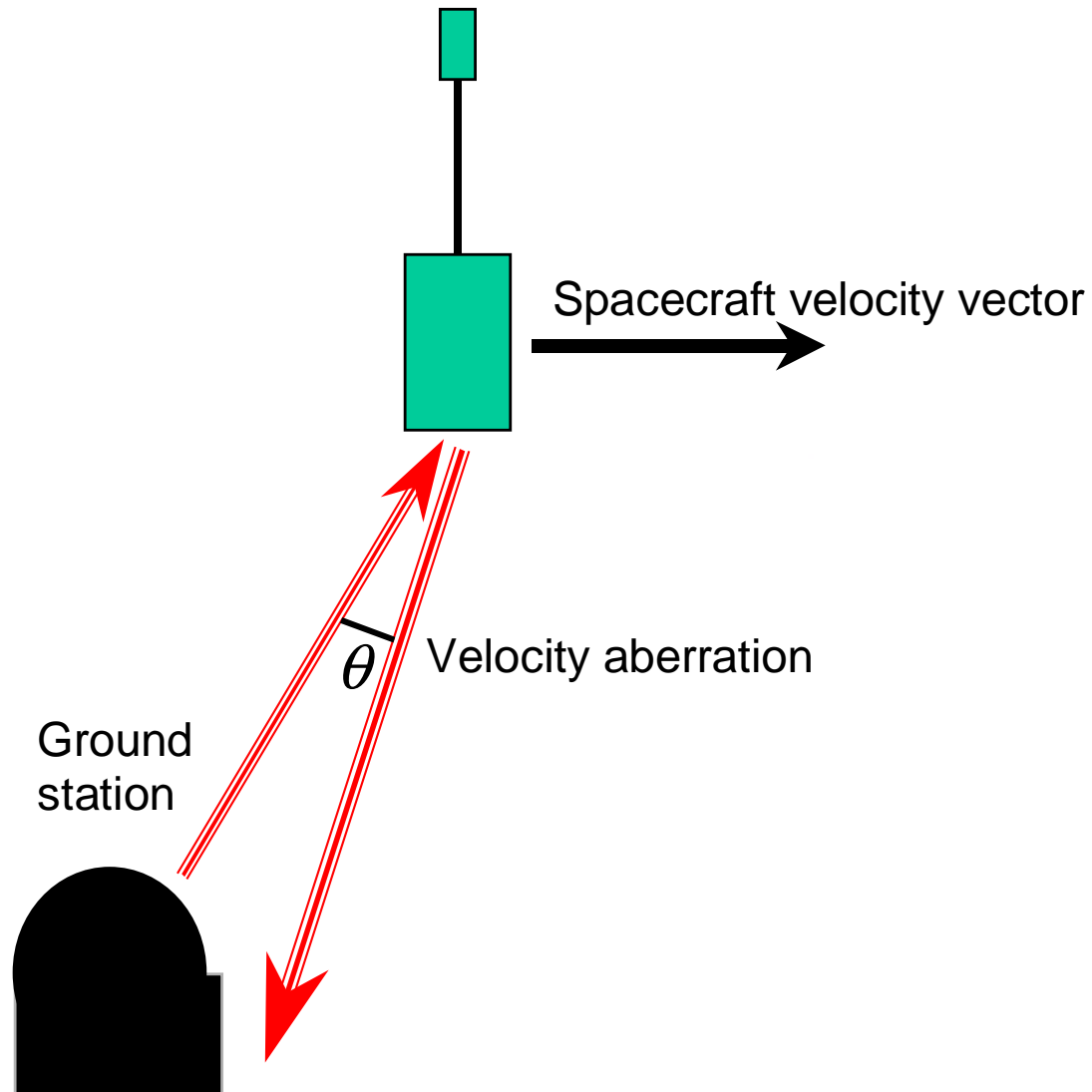


Retroreflector envelope



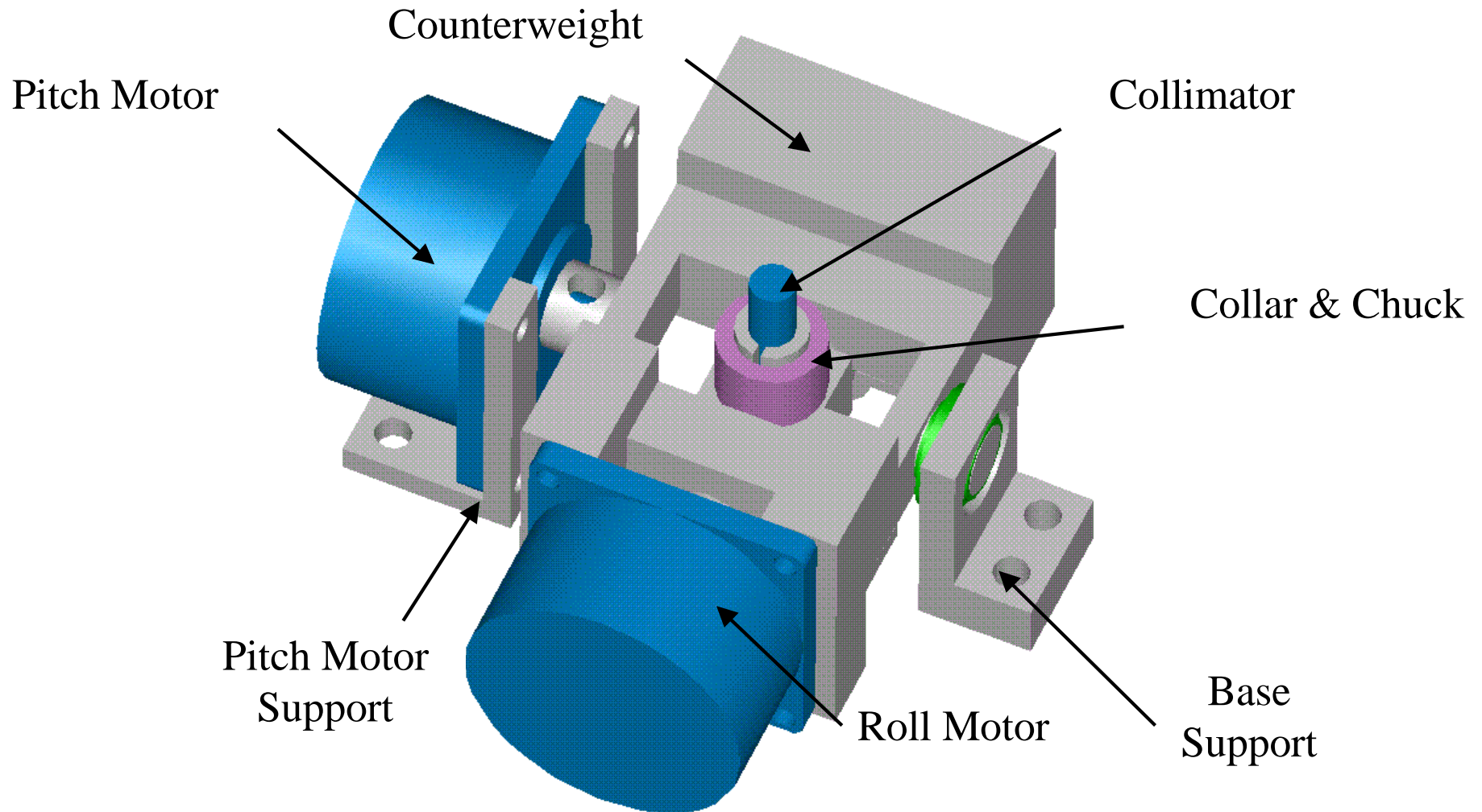
- Maximum dimensions of retroreflector assembly (mirrors and supporting structure)

Velocity Aberration



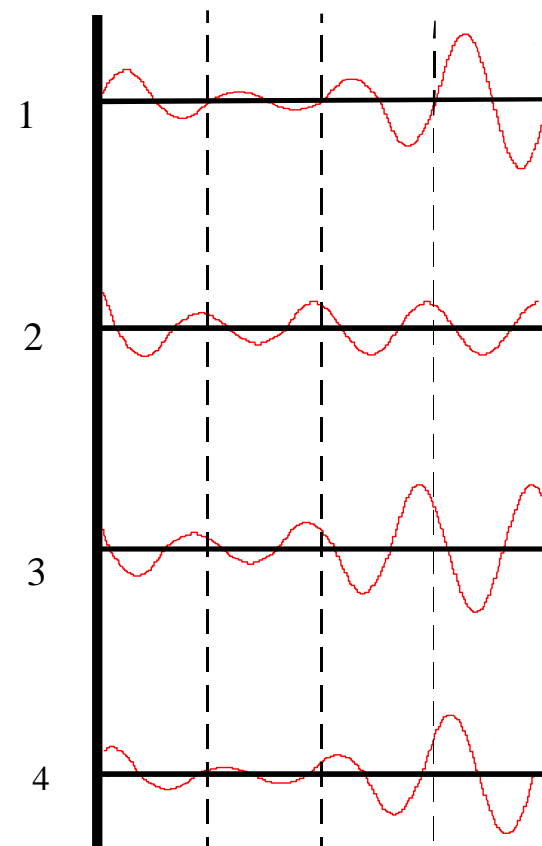
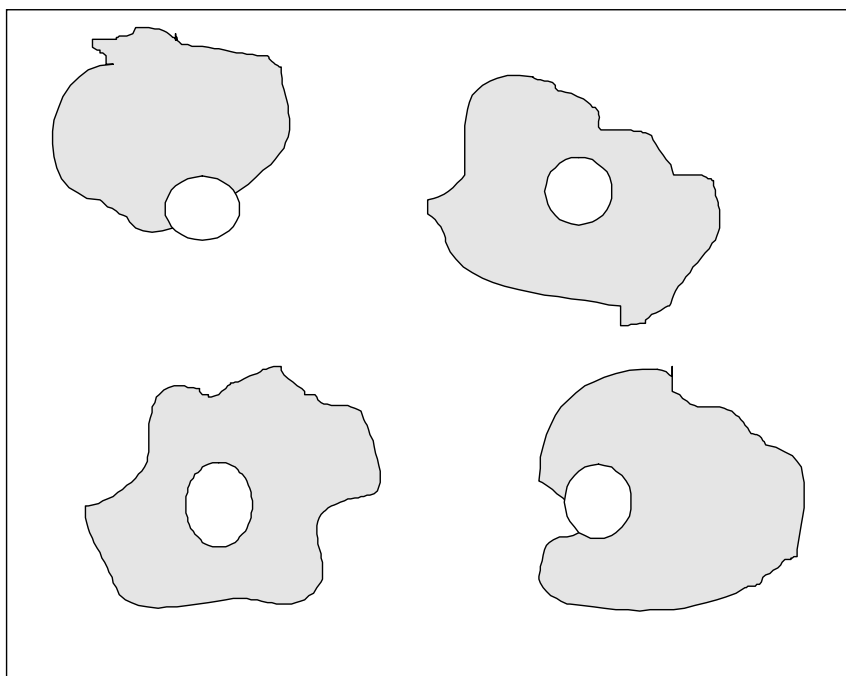
$$\theta = \frac{2v \sin \phi}{c}$$

Gimbaled Beacon

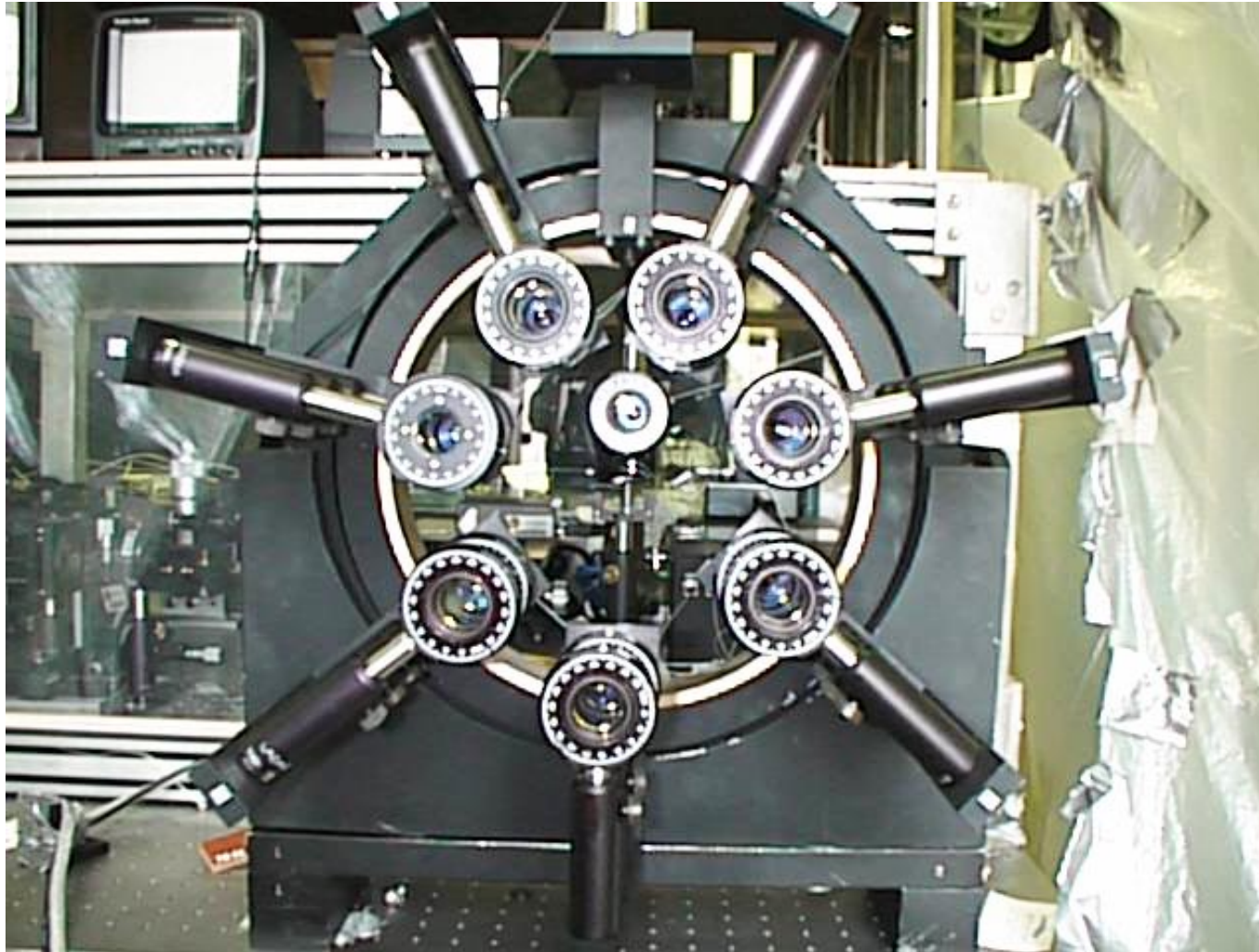




Co-phased array experiment



Eight-element co-phased array receiver





Additional experiments invited



- Contact:

R. Glenn Sellar
Florida Space Institute
12424 Research Parkway, Suite 400
Orlando, FL 32826

407-658-5597
gsellar@mail.ucf.edu